

“CHAI-12U: A Constellation of Schmidt-Dyson Imaging Spectrometers based on the 12U CubeSat Form Factor”

HySpiri Science Symposium June 3rd, 2015

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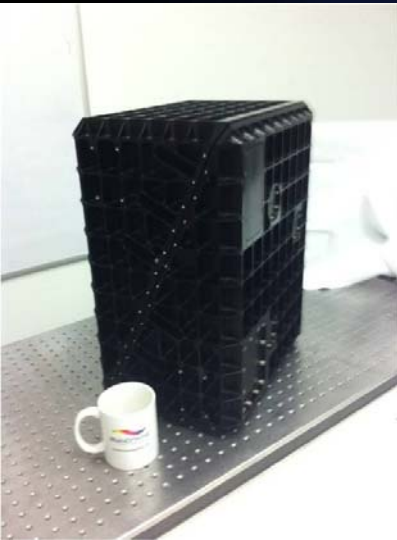
Vightel, Ellicott City MD

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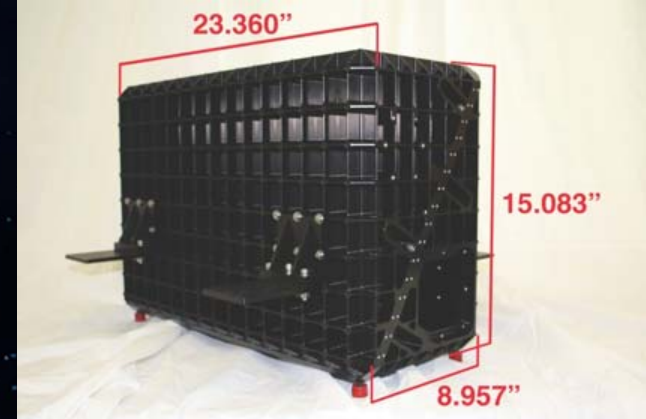
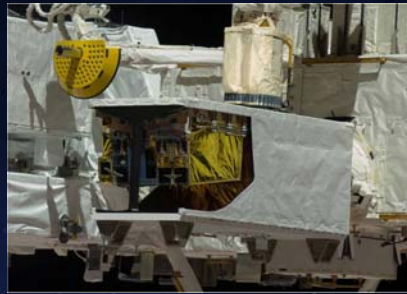
NASA Goddard Space Flight Center

In Memoriam to Robert Lucke, Naval Research Laboratory
as Originator of the Schmidt-Dyson Optical Design





HICO
Spectrograph & Telescope



VIS-SWIR PICARD

CHAI VNIR-2048

BRANDYWINE PHOTONICS



CHAI VNIR-640

CHAI LWIR-320



CHAI SWIR -640



CHAI VNIR-1



CHAI-12U Mission Concept

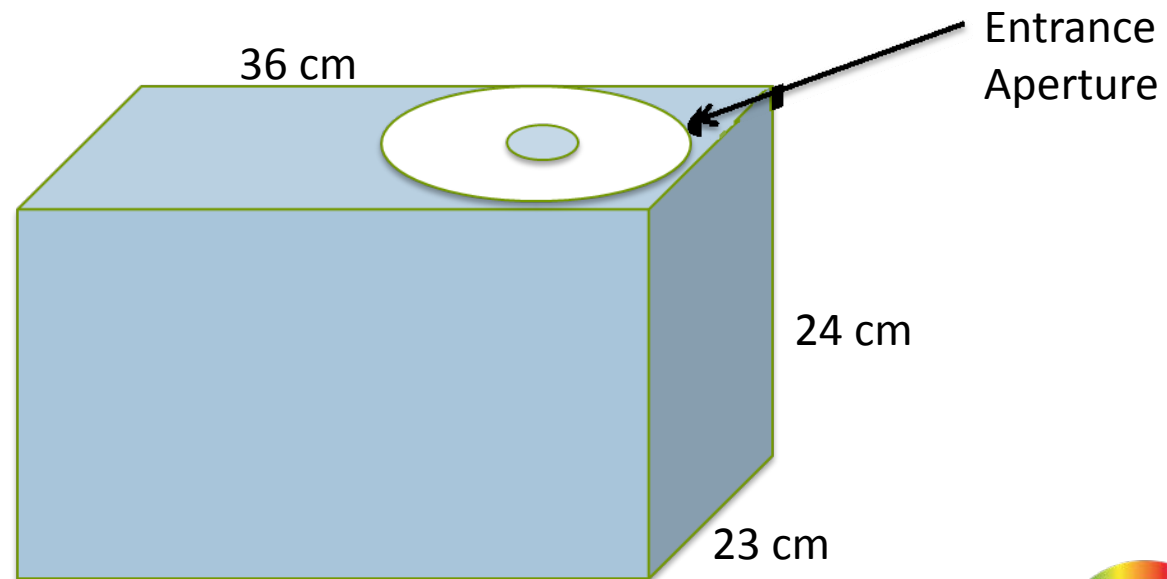
Develop a low-cost constellation of Eight 12U CubeSats to test components, subsystems, algorithms, and mission concepts for addressing HySpiri technology needs and science objectives

Key Advantages of a 12U Constellation

Low-cost, Daily Revisit, Cross-calibration, In-flight Spares, Easier Scheduling, Formation Flying (adjacent, fore/aft for BRDF)
Mission of Opportunity on Hosted Payload or Rideshare, Band selectable. Rapid turn around of data products.

Why 12U CubeSat?

- Enables large aperture without need for deployable telescope
- Easier thermal control than 6U
- Canisters for deployment coming on line within next two years
- Sufficient power for covering VIS-SWIR with cryocooler
- Estimated Cost per Observatory < \$2M to Class-D requirements

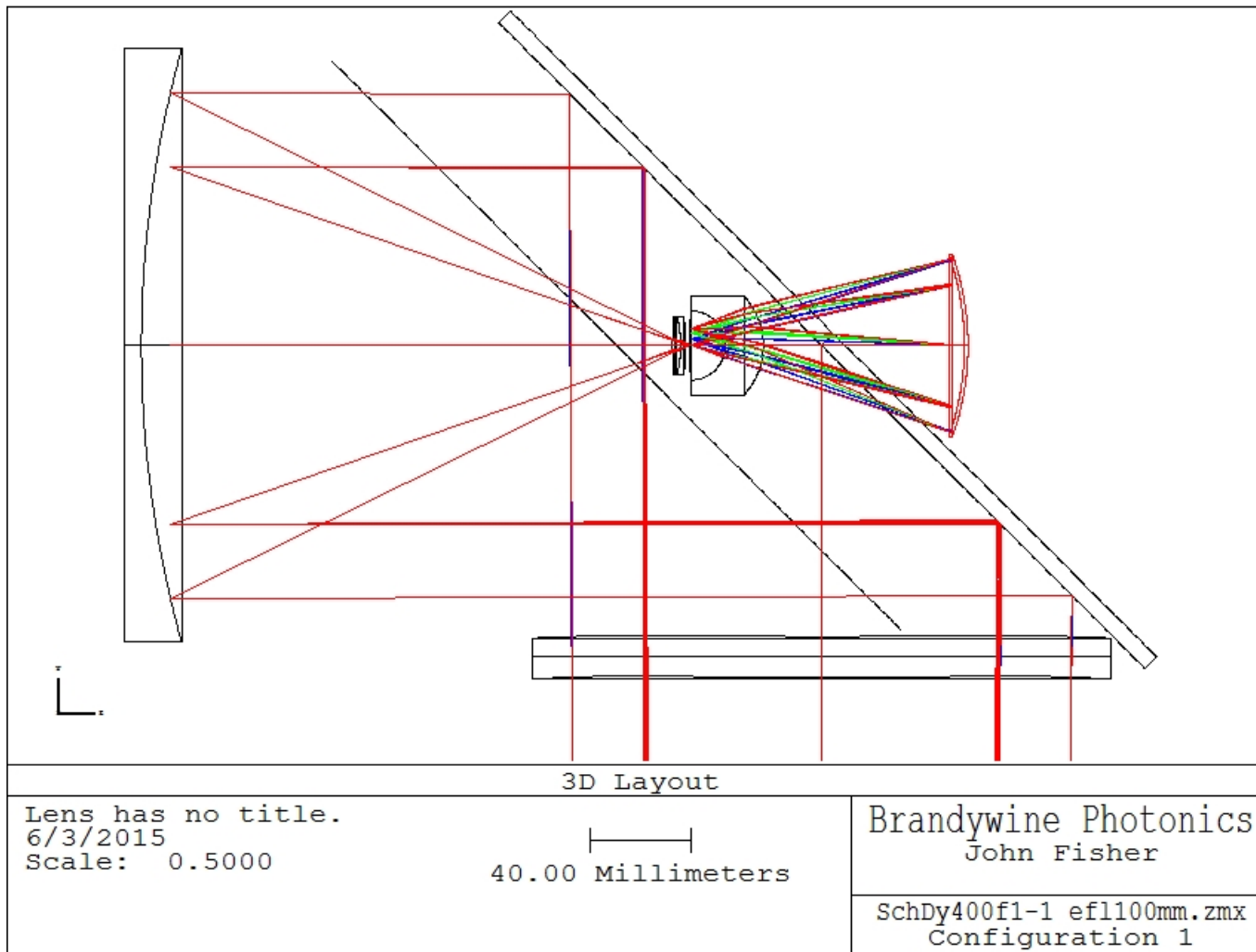


12U CubeSat Format
Mass < 24 kg

CHAI-12U System Specifications

- Focal length: 225 mm, F/1.1, 205 mm aperture, 1500 pixels
- 5.5 degrees FOV, 40 m GSD, 37 km swath width
- Forward Motion Compensation not included in SNR, but could be available (5X improvement in signal)
- Spectral Range 350-1050 nm, 5 nm sampling (140 bands)
- Band selectable by dropping lines during acquisition
- Satellite Estimated SWAP
 - 6 - 10 Kg
 - 25 Watts
 - Fills the 12U form factor 23 cm x 24 cm x 36 cm
 - CHREC 1U Processor (Dan Mandl)
 - X-band transmitter

Optical System Layout

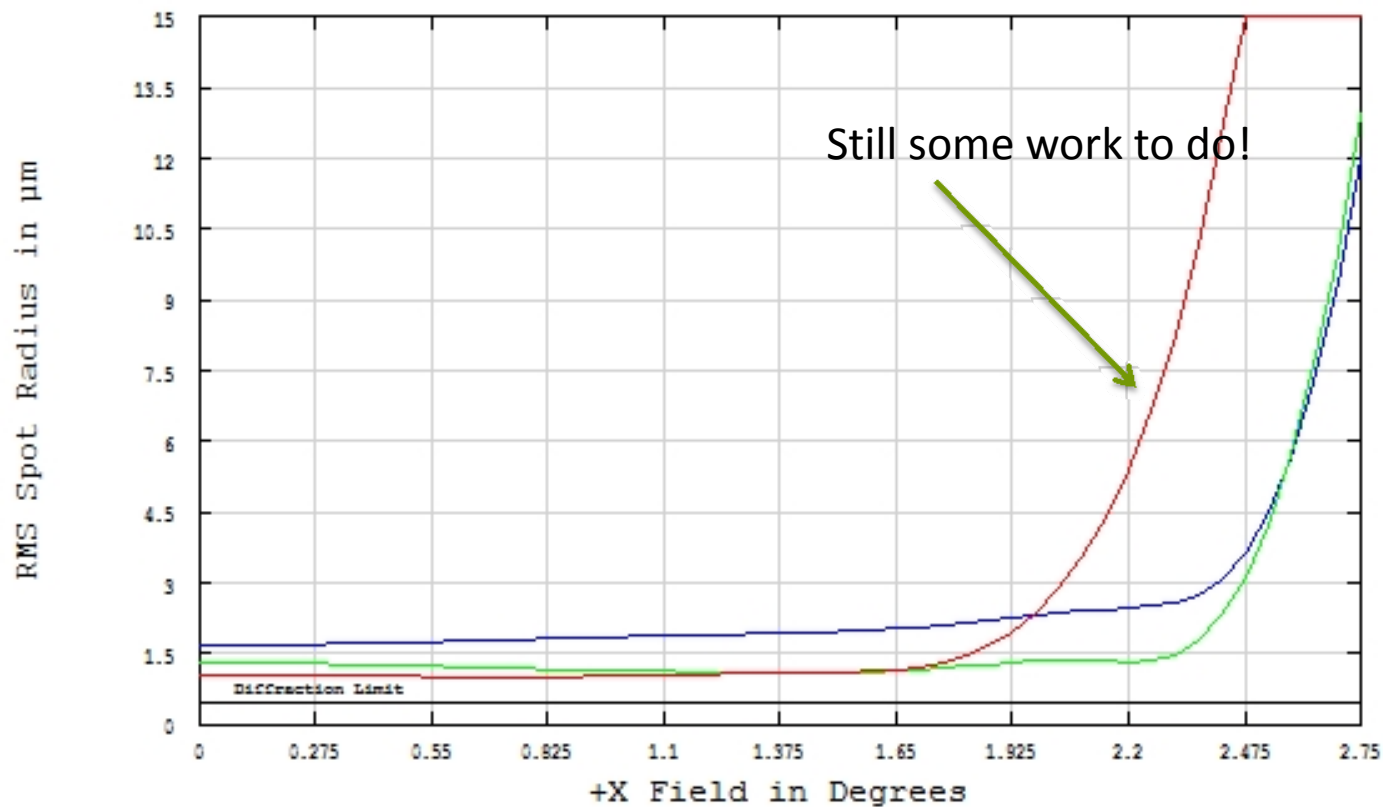


1. Schmidt Corrector
2. Fold Mirror
3. Primary Mirror
4. Corrector lens
5. Slit
6. Dyson Refractive Element
7. Grating
8. FPA

Based on optical design from : R. Lucke and J. Fisher, "The Schmidt-Dyson: a fast space-borne wide-field hyperspectral imager," Proc. SPIE7812, 78120M, 78120M-13 (2010).



RMS Spot Radius vs. Field



RMS Spot Radius vs Field

Lens has no title.

6/2/2015

0.35 0.6 1

Reference: Centroid

Brandywine Photonics

John Fisher
Confidential

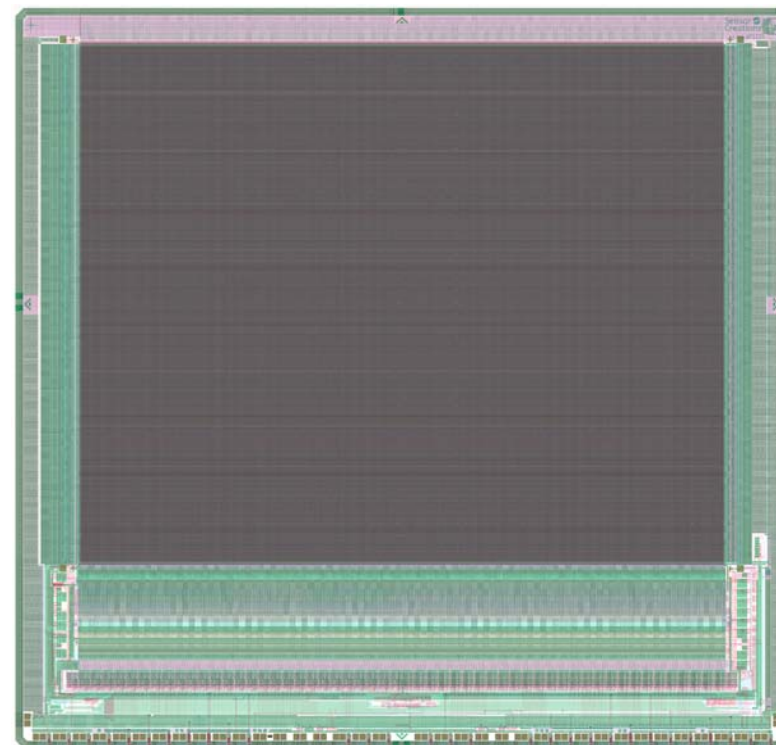
SchDy400f1-1 efl 200mm -jf.ZMX
Configuration 1 of 3

BRANDYWINE
PHOTONICS

FBX – 640x512 CMOS Image Sensor

Summary of design specifications of fully depleted backside illuminated CMOS VGA sensor imager

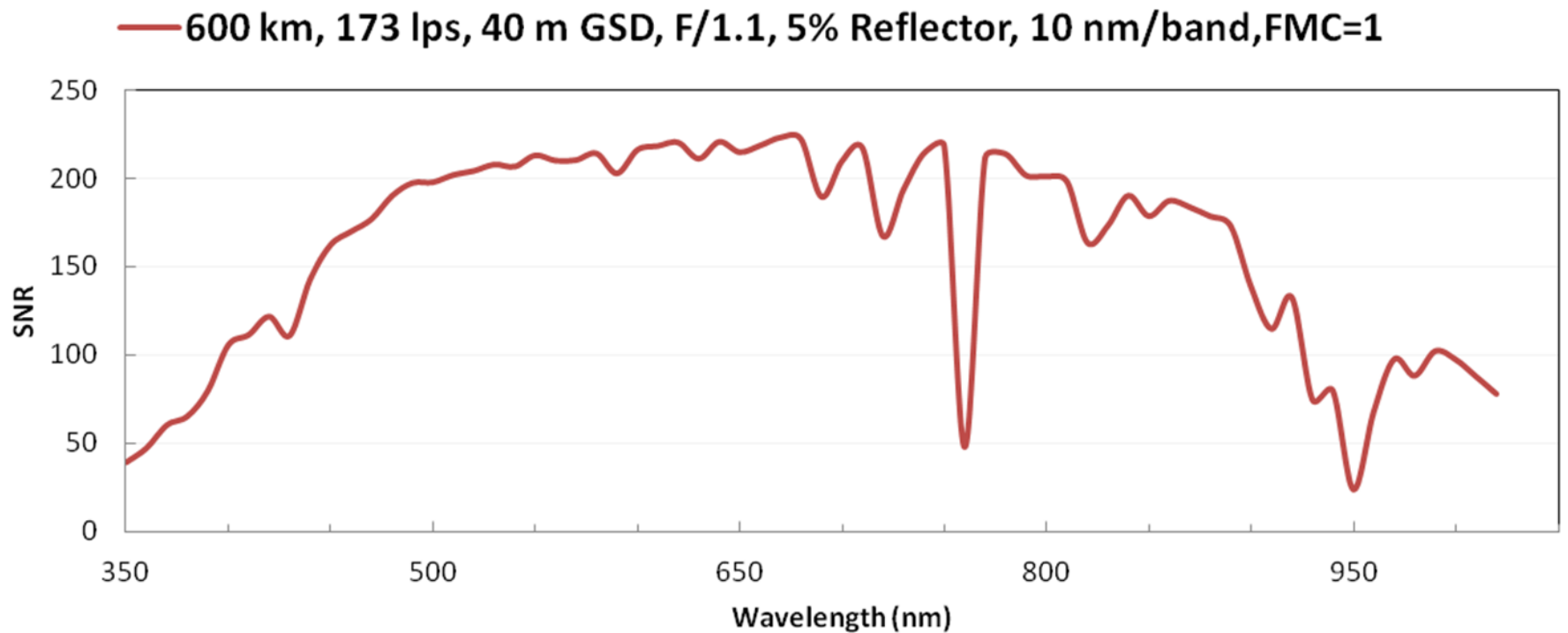
Parameter	Value
Array Format	640 x 512
Pixel Size	15mm
Die Size	12 x 12 mm ²
Frame Rate	30Hz, 60Hz, 500Hz
Shutter type	IWR/ITR/NDR
Exposure Time	200nsec to 30msec
Charge Capacity	500ke, 60ke, 10ke
Minimum Noise	< 10e
QE at 1050nm	> 35% for 300nm thick Si
Binning	2x2, 1x2, 2x1
Master Clock	5 Mhz
Number of analog outputs	1, 2, 4
Power	60 mW @ 30Hz frame rate
Supply Voltages	1.8V/3.3V/10V
Serial Interface	3 wire



*Other High TRL
FPAs available
(Teledyne Chroma)*



SNR 5% Albedo



Increase in SNR performance with FMC=5, higher albedo scenes (vegetation), and wider bands.

CHAI-12U Constellation Advantages

- *Daily revisit, catching major temporal events.*
- *Cross-calibration*
- *Formation Flying (wider swath, 1-hour separation, fore-aft BRDF measurements without nodding)*
- *Greater global coverage*
- *Single launch or satellite failure does not jeopardize the mission*

Thanks to colleagues and NRL, NASA Goddard, and NASA Ames for many fruitful discussions!



System Design Parameters (Back up slide)

Platform

600.0 (km) Altitude
372.8 (mi) Altitude
324.0 (nmi) Altitude
600000 (m) Altitude

circular (type) Orbit
6.38E+06 (m) Approx Radius of Earth
6.98E+06 (m) Radius of Orbit
8.19 (m/s²) Gravitational Acceleration at Altitude
7561.1 (m/s) velocity of satellite perpendicular to ground
5798.7 (s) Orbit Time
96.6 (min) Orbit Time
(m/s) Velocity of Satellite nadir point across earth
6910.9surface

40 (m) Static GSD
172.8 (lines per second)
1 Forward-motion compensation over-sampling factor

Telescope

6.7E-05 (rad) Needed IFOV
66.7 (uRad) Needed IFOV
1.1 (#) F/# of System
0.225 (m) Effective Focal Length
0.2045 (m) Minimum aperture for F-number, signal
8.05 (in) Aperture
5.7 (deg) FOV
0.031 (m) Minimum aperture for spatial resolution

Spectrometer

350 (nm) Shortest Wavelength
2500 (nm) Longest Wavelength
215 (#) Number of Bands
10.00 (nm/band) Dispersion
500 (nm) Blaze Wavelength

Collection

60000 (m) Swath Width (first order)
37.3 (mi) Swath Width
415 (km²/s) Area collection rate
6.69E+08 (bits/s) Data Collection Rate
(Mbytes/s) Data Collection Rate
80(uncompressed)
4.01E+07 (m) Circumference of earth
24,902 (mi) Circumference of earth
(#) Number of orbits to completely cover
668equator
1076 (hours) Time to completely cover equator
45 (days) Time to completely cover equator
(days) Time to completely cover equator - 1/2
90sunlight

Scene

5.0% (refl) Reflectance of Target

Detector

1500 (#) Spatial Channels

Detector

15 (um) Detector Pitch
0.000015 (m) Detector Pitch
0.0058 (s) Integration Time
5 (# e-) Read-out Noise
200000 (# e-) Quantum Well
12 (#) bits per pixel
173 (fps) Frame rate

Title:

600 km, 173 lps, 40 m GSD, F/1.1, 5% Reflector,
10 nm/band,FMC=1

